THE EFFECT OF VALVE OVERLAP ON DETONATION LIMITED ENGINE PERFORMANCE

by

Lieut. Comdr. R. S. McElroy jr, USN
Lieut. Comdr. H. S. Brown, USN
Lieut. Comdr. B. K. Beaver, USN

M. S. Degree June 1947

Thesis
M184

THE EFFECT OF VALVE OVERLAP ON DETONATION-LIMITED ENGINE PERFORMANCE

by

LCDR. R. S. McElroy, Jr., U.S.N.
LCDR. H. S. Brown, Jr., U.S.N.
LCDR. E. K. Beaver, U.S.N.

Submitted in Partial Fulfillment of

the Requirements for the Degree of

Master of Science

in

Aeronautical Engineering

from the

Massachusetts Institute of Technology

1947

Thesis
M184

Cambridge, Massachusetts,
June 1, 1947.

Professor Joseph S. Newell,
Secretary of the Faculty,
Massachusetts Institute of Technology,
Cambridge, Massachusetts.

Dear Sir:

A thesis entitled "THE EFFECT OF VALVE OVERLAP ON DETONATION-LIMITED ENGINE PERFORMANCE" is herewith submitted in partial fulfillment of the requirements for the degree of Master of Science in Aeronautical Engineering.

Respectfully,

ACKNOWLEDGMENTS

The authors wish to express their appreciation to the entire staff of Sloan Laboratories, M.I.T., for their aid in making this thesis possible, and in particular to Professor A. R. Rogowski, Professor W. D. Leary and Mr. J. C. Livengood for their indispensable suggestions and guidance.

TABLE OF CONTENTS

	Page
Introduction	1
Summary of Results	2
Description of Apparatus	4
Procedure	7
Discussion of Results	
80-Octane	9
Di-Iso-Butylene	11
Conclusions	
80-Octane	14
Di-Iso-Butylene	15
Appendix	
Symbols	
Sample Calculations	
References	
Data	
Camshaft Timing Diagrams	
Schematic Diagram	
Rotometer Calibrations	
Performance Curves	
Photographs	

Table of Contents

	Page
Introduction	1
Chapter I: Early Years	3
Immigration of Jewish Farmers	7
Settlements	11
Agricultural Products	14
Co-operation	17
Kibbutz-Moshav Life	21
Conclusion	24
Bibliography	29
Interview Questions	31
Glossary	33
Maps	34
Charts	35
Pictures	36
Newspaper Articles	37
Interviews	38
Bibliography	39

LIST OF FIGURES

Figure No.	Title
1	Camshafts Timing Diagrams.
2	Schematic Diagram Air and Fuel System.
3	Rotometer Calibration Curves.
4	Detonation Limited Inlet Pressure vs R.P.M. (80-Octane).
5	Detonation Limited Inlet Pressure vs R.P.M. (Di-Iso-Butylene).
6	Detonation Limited Inlet Pressure vs Exhaust Pressure.
7	Detonation Limited IMEP vs Exhaust Pressure.
8	Detonation Limited IMEP vs R.P.M.
9	Air Consumption per Cycle vs Valve Overlap (80-Octane).
10	Air Consumption per Cycle vs Valve Overlap (Di-Iso-Butylene).
11	Detonation Limited Inlet Pressure vs Valve Overlap with Constant Exhaust Pressure (Di-Iso-Butylene).
12	Detonation Limited Inlet Pressure vs Valve Overlap with Constant Exhaust Pressure (80-Octane).
13	Detonation Limited Inlet Pressure vs Valve Overlap with Constant R.P.M. (Di-Iso-Butylene).
14	Detonation Limited Inlet Pressure vs Valve Overlap with Constant R.P.M. (80-Octane).
15	Volumetric Efficiency vs R.P.M. (Di-Iso-Butylene).
16	Volumetric Efficiency vs R.P.M. (80-Octane).

Figure No.	Title
1	[illegible] Trading [illegible].
2	[illegible] Virginia City and Truckee [illegible]
3	[illegible] Consolidated Virginia.
4	[illegible] Ophir [illegible] [illegible] by E.T.&C. [illegible].
5	[illegible] Chollar Potosi [illegible] by T.T.&C. [illegible].
6	[illegible] Hale and Norcross [illegible] [illegible].
7	[illegible] Belcher [illegible] [illegible].
8	[illegible] Gould and Curry by T.T.&C.
9	[illegible] Mexican [illegible] by [illegible] [illegible].
10	[illegible] Savage [illegible] by T.T.&C.
11	[illegible] Union [illegible] [illegible] [illegible] [illegible].
12	[illegible] Sierra Nevada [illegible] by T.T.&C. [illegible].
13	[illegible] Ophir [illegible] by T.T.&C. [illegible].
14	[illegible] Gould and Curry by T.T.&C. [illegible].
15	[illegible] Consolidated Virginia by T.T.&C. [illegible].
16	[illegible] Belcher by T.T.&C. [illegible].

INTRODUCTION

The fact that detonation, or knocking, in an internal combustion engine is a function of many related and unrelated variables has caused considerable difficulty in the field of detonation research and engine design. To date, the practice has been to study the effect of these variables one at a time, endeavoring to hold all other variables constant. The purpose of this investigation is to hold all variables constant except the valve overlap and to thereby determine the detonation-limited characteristics of this variable alone.

Four different cam shafts, each with a different valve overlap, have been used in the investigation. The timing diagrams for these shafts are shown in Figure 1. In addition, the complete experiment has been conducted using two different types of fuel: (a) 80 octane automotive, and (b) di-iso-butylene.

Introduction

The first step [illegible], or modeling, in an [illegible] engineering design is a [illegible] study relating [illegible] related variables for [illegible] mathematical similarity in the field of [illegible] research and design [illegible]. In doing the [illegible] are [illegible] to study the effect of [illegible] upon [illegible] alone or in [illegible], endeavoring to hold [illegible] other variables constant. The purpose of this investigation is to hold all variables constant except the ratio covering and to [illegible] determine the [illegible]-fitted characteristics of this variable alone.

Four different [illegible] coils, each with a different ratio overall, have been used in the [illegible]. The finding diagrams for these coils are shown in Figure 1. In addition, the complete experiment has been [illegible] using two different types of [illegible] [illegible] [illegible], and [illegible]-iron-[illegible].

SUMMARY

A series of tests have been conducted on a C.F.R. engine with camshafts whose valve overlap varied from 6° to 120°. In all cases the engine was operated at detonation limited inlet pressure. The tests were performed using two fuels: (a) 80-octane, and (b) di-iso-butylene.

The primary object of the tests was to determine the effect of varying valve overlaps on the detonation limited performance of the engine, and secondarily, to investigate the qualitative effect of residual gas on the fuel characteristics.

In the case of 80-octane, it was found that large overlaps permit the engine to develop greater power at high speed and that residual gas acts as an inhibitor.

In the case of di-iso-butylene, it was found that higher overlaps permit the development of higher power when running at low speeds and that overlap has relatively little effect on the power output at high speed. At the higher speeds the engine detonated at lower critical pressure, as previously discovered by other investigators, but no conclusion can be drawn as to the effect of residual gas on this reversal since, in these tests, this variable could not be isolated.

Abstract

A series of three jets have been analyzed in a $\mathcal{L}/\mathcal{E}$ engine with centrifugal nozzle velocity ratios from ξ_0 to 120^0. In all cases the engine was operated at detonation limited inlet pressure. The tests were performed using two fuels: (a) 80-octane, and (b) di-isopentane.

The primary object of the tests was to determine the effects of varied swirl velocities on the detonation limited performance of the engine, and secondarily, to investigate the qualitative effect of residual gas on the fuel characteristics.

In the case of 80-octane, it was found that increase permit the engine to develop greater speed at high speed and that increased gas ratio is an indicator.

In the case of di-isopentane, it was found that higher overlaps permit the development of higher speed when running at low speeds and that overlap has relatively little effect on the power output at high speed. At the higher speeds the engine exhausted at lower vertical pressure, as previously discussed by other investigators, but an explanation can be given as to the effect of residual gas on this reversal since, in these tests, this variable could not be isolated.

In no case were idling difficulties encountered. However, at least one previous investigator* has found evidence that although a single cylinder engine will idle with high overlap timing, a multi-cylinder engine with similar timing will not. Although this phenomenon was not investigated in this experiment, it is believed to be due to intake and exhaust system dynamic effects.

* Young, A. W., N.A.C.A., A.P.R. (Nov. 1941).

DESCRIPTION OF APPARATUS

ENGINE

A single cylinder, C.F.R., high speed, water-cooled engine having a 3.25-inch bore and a 4.50-inch stroke, was used at a compression ratio of 7 to 1. The engine was directly connected to an electric dynamometer which was employed both as a starter and power absorber for the engine, as well as a motor to turn the engine over while measuring friction torque. A shrouded intake valve was used to enhance distribution and mixing of the fresh charge in the cylinder and an Auto Lite B-5 spark plug was used to mitigate any tendencies toward pre-ignition.

FUEL SYSTEM

The fuel system is shown schematically in Figure 2. The fuel was taken from the fuel tank through a small electrically driven pump to a bubble separator and surge tank. Compressed air was pumped into the bubble separator and the pressure was held constant at 30 psi. This air acted as a "cushion" for the system. From the bubble separator the fuel was passed through one of two Fisher and Porter rotometers which had been calibrated by the authors prior to the experiment, as shown by Figure 3. From the rotometers the fuel flowed directly into the vaporizing tank.

AIR SYSTEM

The air system is shown schematically in Figure 2.

DESCRIPTION OF APPARATUS

Engine

A single cylinder, 4-H.P., light opposed, motor-cycle engine having a 3.25-inch bore and a 3.50-inch stroke, was used at a compression ratio of 7 to 1. The engine was directly connected to an electric dynamometer which was employed both as a source for power absorber for the engine, as well as a motor to turn the engine over while measuring friction horsepower. A motorized intake valve was used to synchronize distribution and timing of the fuel charge to the cylinder and an auto tire 3-½ square plug was used to ignite any hydrocarbons burned pre-ignition.

Fuel System

The fuel system is shown schematically in Figure 2. The fuel was taken from the fuel tank through a small electrically driven pump to a variable regulator and surge tank. Compressed air was pumped into the surge reservoir and the pressure was held constant at 10 psi. This air worked as a "cushion" for the system. From the surge chamber the fuel was passed through one of two flow-rate meters selected for the operations, as shown by Figure 3. From the meters the fuel flowed directly into the vaporizing tank.

Air System

The air system is shown schematically in Figure 2.

Supercharged air was tapped directly from the laboratory main. It was then taken through two pressure regulators and metered through one of two flat plate orifices with flange taps. The smaller of the two orifices was .515 inches in diameter and the larger was .725 inches in diameter. Both orifices were constructed according to ASME specifications and inserted in 2-inch pipes. After being metered, the air passed through double surge tanks into the vaporizing tank.

VAPORIZING TANK

Air and liquid fuel were introduced into the vaporizing tank separately. The fuel was here vaporized and mixed with the air. The mixture was held at a constant temperature of 140°F. This temperature was controlled by the circulation of steam through heating coils within the tank. From the tank the fresh mixture passed through the throttle valve and into the engine.

OIL SYSTEM AND WATER JACKET SYSTEM

The jacket water temperature was held constant at 212°F. and was controlled by circulating water through a cooler between the condenser and the jacket. The oil temperature was held constant at 140°F. by means of circulating steam or cooling water through the two heat exchangers.

MEASURING INSTRUMENTS

The engine power output was absorbed by a Star electric dynamometer and the brake torque was balanced by a hydraulic

piston attached through a lever arm to the dynamometer casing. The oil pressure on the hydraulic system was measured in inches of mercury in an open end manometer. The hydraulic system was designed so that one inch of mercury was the equivalent of a one-pound force acting on the hydraulic piston. The reading of the dynamometer scale in inches of mercury was converted to pounds per square inch, brake mean effective pressure. The electric power output from the dynamometer was dissipated through a grid at low outputs and at higher outputs the power was pumped into 110-volt or 220-volt mains, depending upon the magnitude of the power generated.

Detonation was measured qualitatively by employing a detonation pickup in conjunction with a cathode ray oscilloscope. The pickup, a Draper flat diaphragm type with a natural frequency of about 95,000 cycles, was inserted in the cylinder head. The output from this pickup was analyzed and projected on the screen of the oscilloscope as a curve of relative rate of change of cylinder pressure vs time. When conditions of insipient detonation were reached, the curve displayed a definite break or "pip". The operator observing the "pip" also controlled the throttle.

Engine speed was roughly measured by using a tachometer, and for finer speed adjustment a strobotac was employed. Speed was controlled by a drop wire resistance in the shunt field of the dynamometer.

PROCEDURE

The following table indicates the operating conditions which were held constant throughout the entire experiment:

Compression ratio	7.0
Fuel-Air ratio	0.08
Spark advance	18° B.T.C.
Oil temperature	140° F.
Water jacket temperature	212° F.
Inlet mixture temperature	140° F.

The following quantities were varied:

Valve overlap	6° - 120°
Manifold pressure	20 - 80 " H_g
RPM	1000 - 3000
Exhaust pressure	30 - 6 " Hg

Each camshaft (with valve timings as per Figure 1) was in turn installed in the engine and two tests were performed. Test no. 1 consisted of varying the R.P.M. and running up the manifold pressure until insipient detonation was reached. Test no. 2 consisted of operating the engine at constant R.P.M. (2000), steadily reducing the exhaust back pressure, and then running up the manifold pressure until insipient detonation was reached. Both of these tests were performed using two fuels (80-octane and di-iso-butylene).

As the insipient detonation points were approached, adjustment of the fuel-air ratio was obtained by computing

Procedure

The following table indicates the operating conditions which were held constant throughout the entire experiment:

Compression ratio	7:1
Feed-air ratio	0.66
Spark advance	18° B.T.C.
Oil temperature	150° F.
Water jacket temperature	212° F.
Inlet mixture temperature	110° F.

The following quantities were tested:

Inlet pressure	0° - 180°
Quantity jacketed	30 - 80 = 50
RPM	1500 - 3000
Exhaust pressure	30 - 0 = 0g

Each somewhat (with minor changes as per Figure 1) was run located in the engine and then tests were performed. Part two. I consisted of varying the R.P.M. and keeping at the specific pressure until inspected fluctuation was reached. Part two. I consisted of operating the engine at constant R.P.M. (3000), steadily reducing the exhaust back pressure, and then reading up the specific pressure until inspected deflection was reached. Both of these tests were performed using two fuels (80-octane and 91-low-octane).

As the inspiant exhaust valve was eliminated, adjustment of the feed-air ratio was obtained by operating

the mass of air flow by the orifice equation, entering the rotometer calibration chart with the mass of fuel flow desired to give a fuel-air ratio of .08, and adjusting the fuel flow until the rotometer showed the proper scale reading. The manifold pressure was then varied until insipient detonation occurred. The above procedure was repeated until a "no change" condition was reached.

DISCUSSION (SO FAR)

From Fig. 6 it is noted that as the residual gas is "sucked" out of the cylinder due to decreasing the back pressure, the allowable Pi is progressively lowered in the case of the 6° overlap cam, and it is therefore apparent that residual gas has an inhibiting effect on detonation.

Still referring to Fig. 6, it might reasonably follow that if this is true, the higher overlaps should give even less detonation limited Pi. However, in the case of the higher overlaps, more fresh mixture is consumed per cycle, tending to cool the engine parts. This increased cooling may overshadow the effect of reduced residual gas and thereby permit higher allowable Pi's.

Referring to Fig. 4, it is noted that at low speed (1000 RPM), higher overlaps permit higher allowable Pi's. At this speed the inlet pressure is always less than the exhaust pressure. Since the mixture consumption per cycle is approximately constant (at 1000 RPM), as seen by Fig. 9, differences in engine parts temperature due to this effect are negligible. However, two other effects influence the amount of residual back flow, (a) the pressure gradient between Pe and Pi, and (b) the overlap which changes the time available for the back flow to take place. If it is assumed that residual gas acts as an inhibitor, it follows that at 1000 RPM the residual gas content is progressively higher with greater overlaps. As the engine speed is in-

creased, both the pressure gradient and the time available for scavenging are reduced, and at 3000 RPM all overlaps show approximately the same detonation limited Pi's.

Fig. 8 indicates that overlap has little effect on detonation limited IMP at low speed, but that overlap increases the IMP appreciably at 3000 RPM. It is to be noted that at this speed the highest overlap gives the maximum power, but that a reduction in overlap of 50% reduces the power by only 7%.

At 2000 RPM, Fig. 7 indicates that for all overlaps greater than 60°, the variation of IMP with altitude is negligible. If these runs had been conducted at a lower speed, the overlap effect might have become more pronounced due to greater time available for purging, or back flow, as the case might be.

Fig. 8 indicates that low overlaps have the better volumetric efficiencies at low speed, but that a reversal takes place such that the high overlaps give the better volumetric efficiencies at high speed. This reversal is due to the fact that at low speed Pi is less than Pe, while at high speed Pi is greater than Pe (Fig. 4).

DISCUSSION (DI-ISO-BUTYLENE)

In the case of di-iso-butylene, the determination of insipient detonation was exceedingly more difficult than in the case of 80-octane. During the early stages of operation, the authors were somewhat deceived by what was observed on the oscilloscope. When the inlet pressure was increased to the point where a distinct "pip" was observed on the scope, it was assumed that insipient detonation was reached in that the pip resembled that obtained when insipient detonation occurred with 80-octane. However, it was later discovered that after the initial "pip" appeared, its intensity was unaltered by either radical changes in spark timing or large increases of manifold pressure (in the order of magnitude of 12 to 18 inches of Hg). After this discovery, the initial "pip" was disregarded and insipient detonation was only assumed to be present when an increase in manifold pressure noticeably increased the "pip's" magnitude. This fact necessitated a re-run of the 6° overlap shaft on di-iso-butylene for varying MAP's and the re-run was performed at the conclusion of the other runs. However, the new data was erringly placed in table I and also appears in figure 13 for the low overlap. These four points should be raised by about 15" Hg.

The running of di-iso-butylene in addition to 80-octane was undertaken to determine if the commonly observed reversal of critical pressure-speed relations, as compared with 80-octane, would persist in the case of large variations in

Discussion (Tetrachloroethylene)

In this case of tetrachloroethylene, the determination of hepatitis infectivity was successfully carried out initially when in the case of chloroform. During the early stages of examination, the patients were routinely observed to find any abnormal on the conditioning. When the first patient was discovered to have hepatitis during a routine "pig" not examined on the scope, it was assumed that laboratory contamination had occurred in that the pig accession that morning had been taken directly from the cage. However, it was later discovered that after the initial "pig" injected, its intensity was monitored by clinical chemistry in order finding or large increases of specific enzymes (in the order of magnitude of 10 to 1000 times). Other possible diseases, the initial "pig" was discovered and laparotomy examination was done because on the personal death as apparent infection was both necessary to be proved with laparotomy in hepatitis previous positively increased the "pig's" enzymes. This test necessitated a re-use of the CO_2 storage cavity on tetrachloroethylene for another study and the re-use now performed at the evacuation of the other tubes. However, the new data was strikingly similar to table 2 and what appears in figure 3) for the time overall. These test points should be joined by about 15° CO.

The toxicity of tetrachloroethylene in addition to the above mentioned to determine if the commonly observed presence of artificial pressure-caused reactions, is assisted with the caution, should persist in the case of large variations in

residual gas concentration. It was suspected that residual gases might exert either a catalytic or an inhibiting effect on the auto ignition of the last part of the charge to burn. At the time of this investigation, the effect of residual content on time reversal was being concurrently studied by Lieutenant Commander J. M. Richards, et al, under carefully controlled experimental conditions and it was anticipated that the two investigations would perhaps substantiate each other through the viewpoint of different experimental parameters. The work of Richards, et al, showed conclusively that residual gas has no appreciable effect either as a catalyst or as an inhibitor on the observed reversal. Although the investigation presented herein also shows the reversal (Fig. 10), no theory can be propounded by this experiment in view of the fact that residual gas effects are negligible. Sloan Laboratories, M.I.T., have shown the reversal to be due to a time effect (rapidity of compression).

It is believed by the authors that for similar investigations in the future, better results might be obtained by increasing the inlet temperature of the fresh mixture above 140°F since this fuel is relatively insensitive to pressure.

As observed from Figs. 11 and 12, the effect of overlap on detonation limited Pi is negligible with a slight tendency toward higher Pi's for higher overlap in the case of 80-octane and lower Pi's for the high overlaps with di-iso-butylene.

The IHP is progressively greater for higher overlaps at 1000-1100 RPM, and at increasing speeds the effect of overlap

[illegible] can contribution. It was observed that reticulocyte count after a reduction in the initial concentration of the anti-Ig portion of the line were at the higher [illegible] reticulocyte formation, the effect of reticulocyte counts on blood reservoir was later reconstructed [illegible] Laboratory Communication 1, D. Pickering, et al. showed essentially complete suppressive conditions and it was postulated that the [illegible] might possibly stimulate each other through the elaboration of different suppressor products. The work of Osborne, et al. showed conclusively that reticulocytes can be suppressed either as a complex or in the initial or the abnormal interval. Although the investigation presented herein also shows the interval (Fig. 3c), the effect of this suppression in view of the fact that reticulocyte counts are negligible. Other investigators, e.g., have shown the reticulocyte to be due to a slow uptake (rapidity of suppression).

It is believed by the authors that the clinical investigations in the future, better results might be obtained by increasing the later interpretation of the final clinical study. IgG[illegible] alone only had a relatively insensitive to immunosuppression observed from Figs. 2 and 3, the effect of reticulocyte depression limited it is negligible with a slight increase toward higher IgG's for higher overlap in the case of IgG-coated and lower IgG's for the high overlaps with Ig-coated.

The IgG is progressively greater the higher reticulocyte IgG, and at increasing amounts the effect of reticulocyte

becomes less important since the IMEP is substantially constant (Fig. 6). It is believed certain that the 120° overlap point at 2100 RPM (Fig. 6) is considerably in error and that the IMEP for this point should be about 185, as shown by Fig. 7 at Pe = 30 and RPM = 2000. The error is unquestionably due to a personnel oversight. Time permitting, this shaft would be inserted in the engine and the point recomputed.

Fig. 15 indicates extensive blow through for the high overlap cam at 1100 RPM.

CONCLUSIONS (ISO-OCTANE)

For maximum detonation limited IMP optimum valve overlap depends upon engine speed.

Overlap has little effect on detonation limited IMP at low speed.

Higher overlaps give higher allowable IMP's at high speed.

No appreciable gain is obtained in the indicated output for overlaps greater than 60°.

At a speed of 2000 RPM the indicated output changes negligibly with decreases in back pressure for all overlaps of 60° or greater.

Residual gas has an inhibiting effect on the detonation tendencies of this fuel.

It has previously been established that residual gas acts as an inhibitor in the case of iso-octane fuel.*

* N.A.C.A., T.R. #699 (1940).

CONCLUSIONS (II-I.O-OCTYLENE)

This fuel, as compared with 80-octane, permits higher detonation limited inlet pressure and higher engine output under all conditions investigated.

This fuel is relatively insensitive to pressure.

As the speed is increased, the allowable fresh mixture consumption per cycle is reduced, resulting in the peak pressure reversal observed by previous investigators.

No conclusion is drawn as to the effect of residual gas on the detonation characteristics of this fuel.

The high allowable Pi at low speeds causes excessive "blow through" when using the 120° overlap cam.

Valve overlap has relatively little effect on detonation limited inlet pressure.

Greater overlaps permit the development of higher IMP's at low speed, but as the speed is increased, the IMP is substantially constant for all overlaps.

SYMBOLS

B.L.	Brake loading	"Hg.
e	Volumetric efficiency	-
FL	Friction loading	"Hg.
F/A	Fuel air ratio	-
Fuel Cons.	Fuel consumption	#/sec.
Imep	Indicated mean effective pressure	Psia.
IHP	Indicated horse power	H.P.
Isfc	Indicated specific fuel consumption	$\frac{\#\ \text{fuel}}{\text{IHP-hr.}}$
Ma	Air consumption	#/sec.
O	Orifice in use	-
ΔP	Pressure across air orifice	$"H_2O$
Pb	Pressure before air orifice (abs.)	"Hg.
Pe	Exhaust manifold pressure (abs.)	"Hg.
Pi	Inlet manifold pressure (abs.)	"Hg.
R	Rotometer in use	-
Roto	Rotometer reading	-
Tb	Temp. before air orifice	^oF
Vd	Engine displacement volume (37.33 cu.in.)	-
ρi	Inlet mixture density	$\frac{lb}{ft^3}$

Symbols

EQUATIONS

For .515" orifice:

$$\text{Air consumption} = .01625 \left(\frac{P_b \times \Delta P}{T_b + 460}\right)^{\frac{1}{2}}$$

For .725" orifice:

$$\text{Air consumption} = .0365 \left(\frac{P_b \times \Delta P}{T_b + 460}\right)^{\frac{1}{2}}$$

Fuel consumption = (0.08) x Air consumption

$$\text{Imep} = (BL + FL)(4.245)$$

$$\text{IHP} = \frac{(BL + FL)(RPM)}{5000}$$

$$\text{Isfc} = \frac{\text{\# fuel/hr}}{\text{IHP}}$$

$$e = \frac{(\text{Const.})M_a}{\rho_1 \times Vd \times RPM}$$

REFERENCES

1. Taylor, C. F. and Taylor, E. S., "The Internal Combustion Engine".
2. Pratt & Whitney Memorandum Report #492.
3. Livengood, J. C. and Eppes, J. V. D., "The Effect of Changing Manifold Pressure, Exhaust Pressure, and Valve Timing on the Capacity and Output of a Four Stroke Engine Operated with Inlet Valves of Various Diameters and Lifts", (April 1945).
4. Glover, J. H., "The Effect of Valve Overlap on the Performance of an Engine with Inlet Pressure Higher than Exhaust Pressure".
5. Barkleroad, N. E. and Gill, P. W., "Investigation of Detonation Characteristics of Various Fuels", (June 1946).
6. NACA, A.P.R., November 1941 (A. W. Young).

References

1. Taylor, D. F. and Taylor, F. S., "The Internal Combustion Engine".

2. Diesel & Gibbons Technical Report 1977.

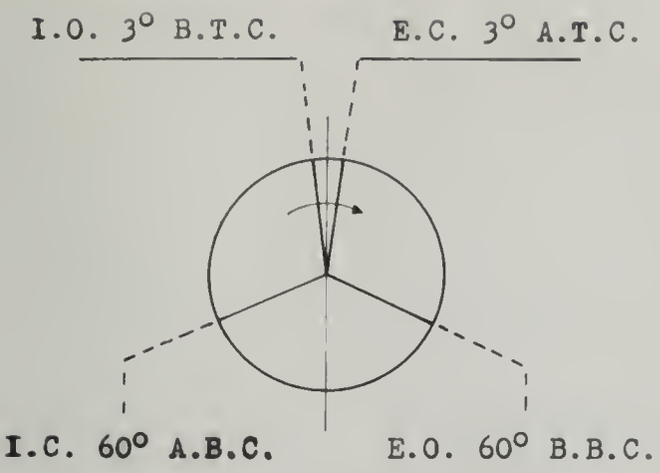
3. Ellenwood, V. D. and Lyons, E. T. S., "The Effect of Chemical Reactive Pressure, Volume and Time on the Power and Duty Cycle of a Four Stroke Cycle Operation with Twin Spark Plugs and Varied Clearance and Load", August 1940).

4. Olsson, J. Ry, "The Effect of Water Cooling on the Performance of an Engine with Light Pressure Steam Heat Element Present".

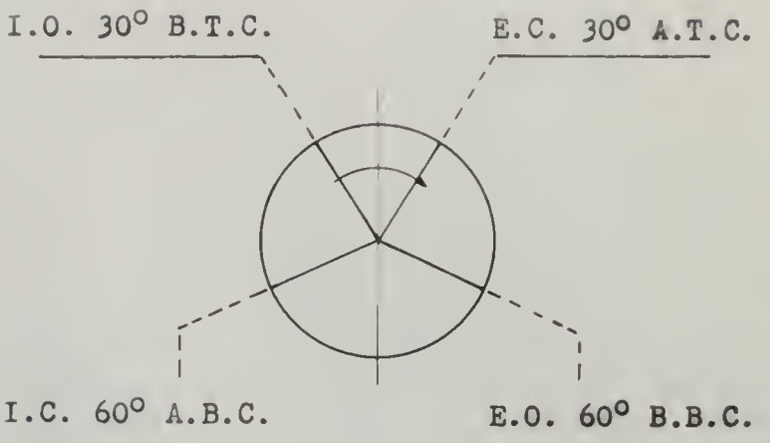
5. Barkley, R. E. and Ellis, F. W., "Thermodynamics of Combustion Characteristics of Various Fuels", (June 1950).

6. Mich, E.J.E., Thermodynamic Cycle (4, Temp).

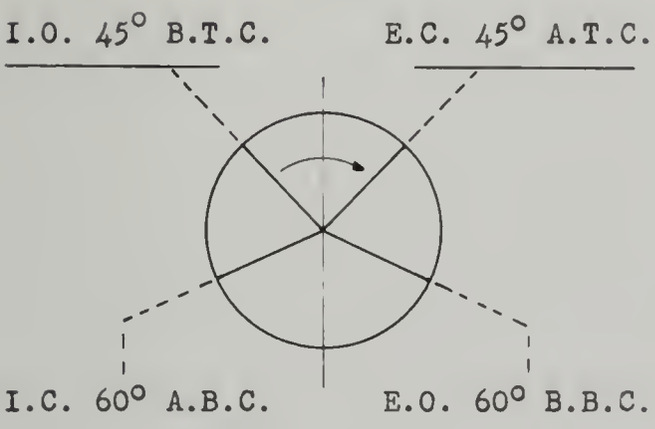
CAM NO. 1



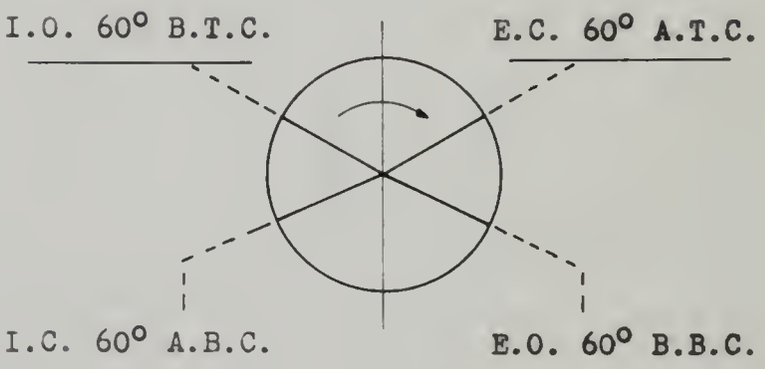
CAM NO. 2



CAM NO. 3

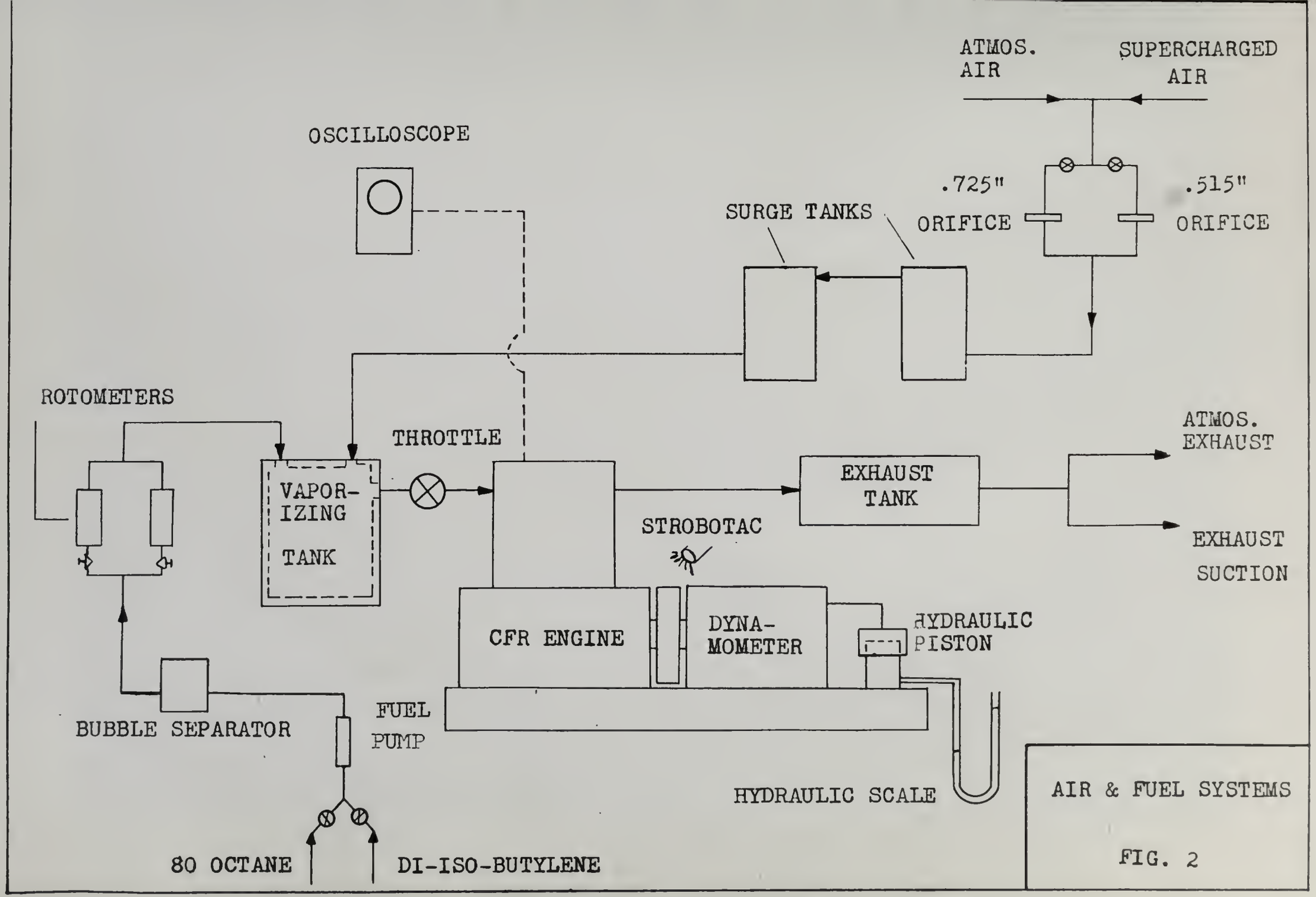


CAM NO. 4



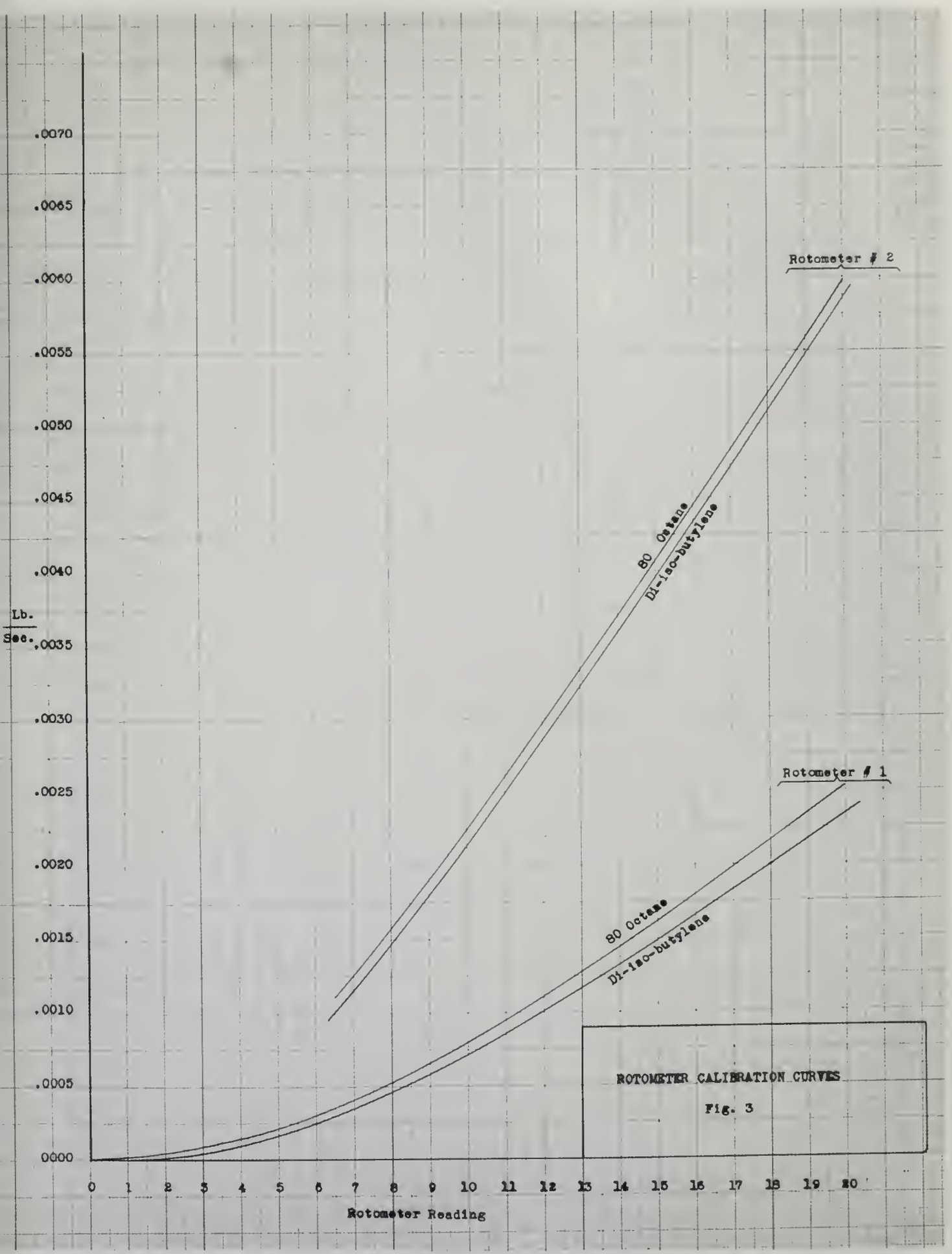
CAMSHAFT TIMING DIAGRAMS

FIG. 1



AIR & FUEL SYSTEMS

FIG. 2



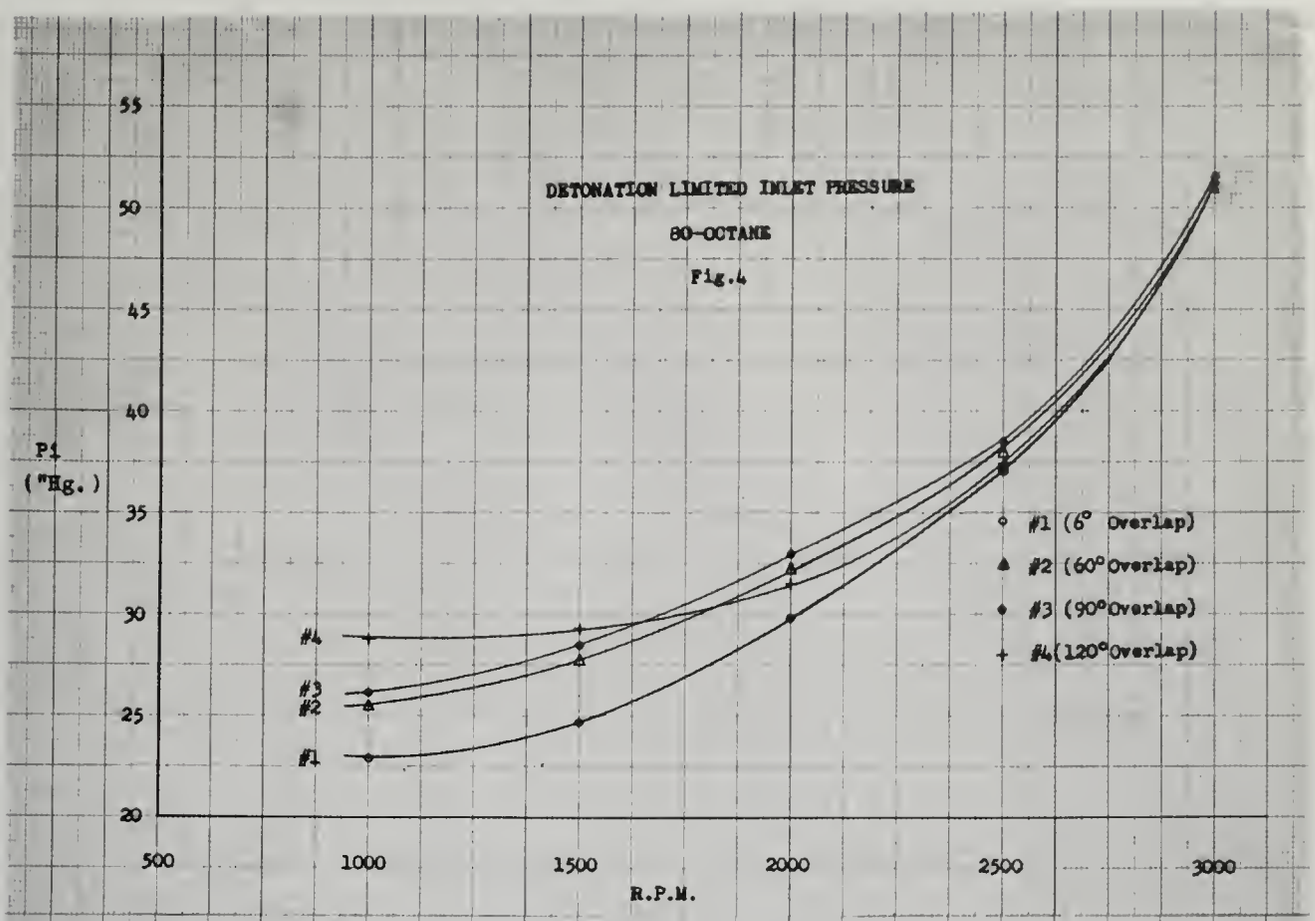
ROTOMETER CALIBRATION CURVES

Fig. 3

DETONATION LIMITED INLET PRESSURE

80-OCTANE

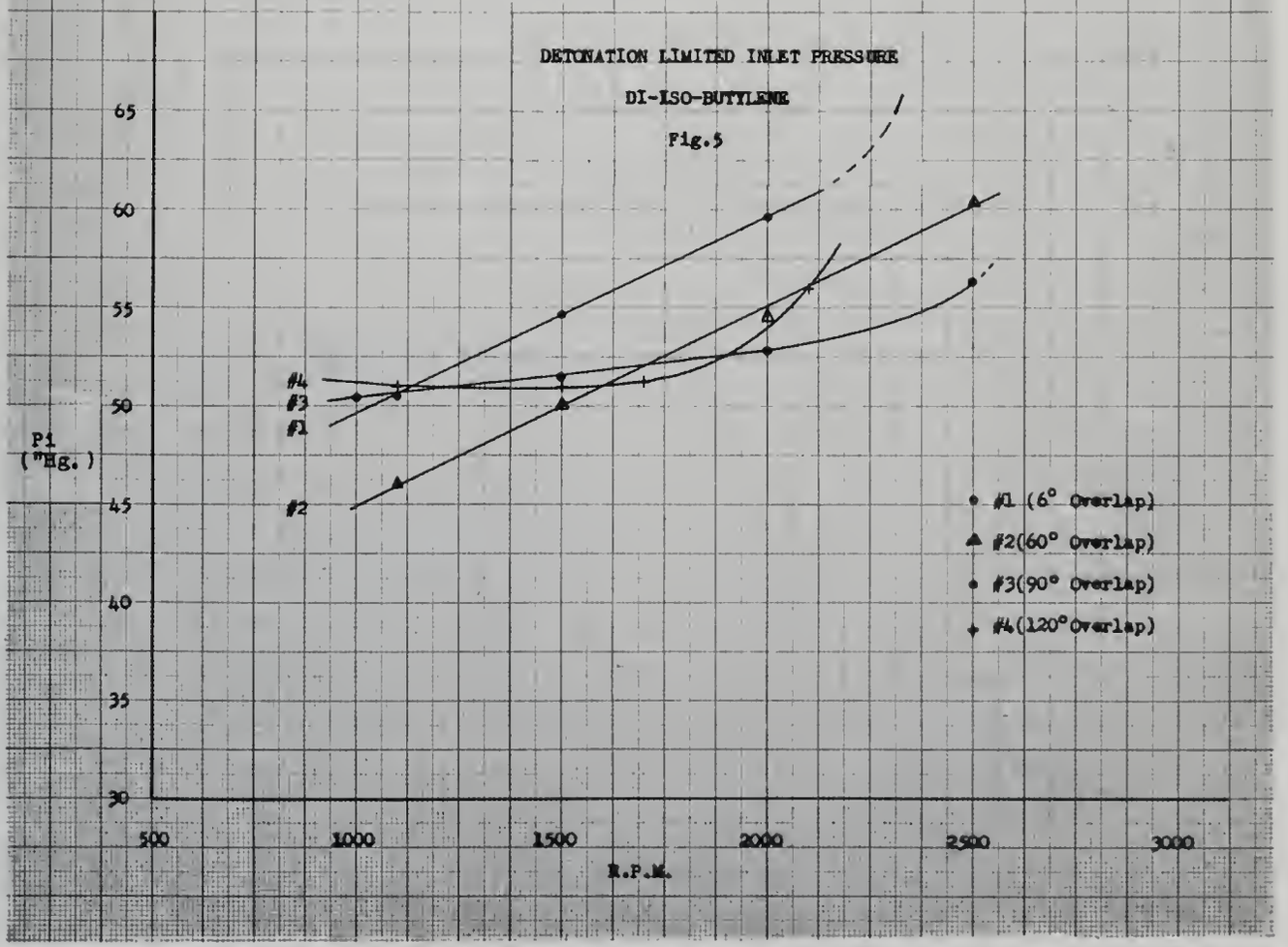
Fig.4



DETONATION LIMITED INLET PRESSURE

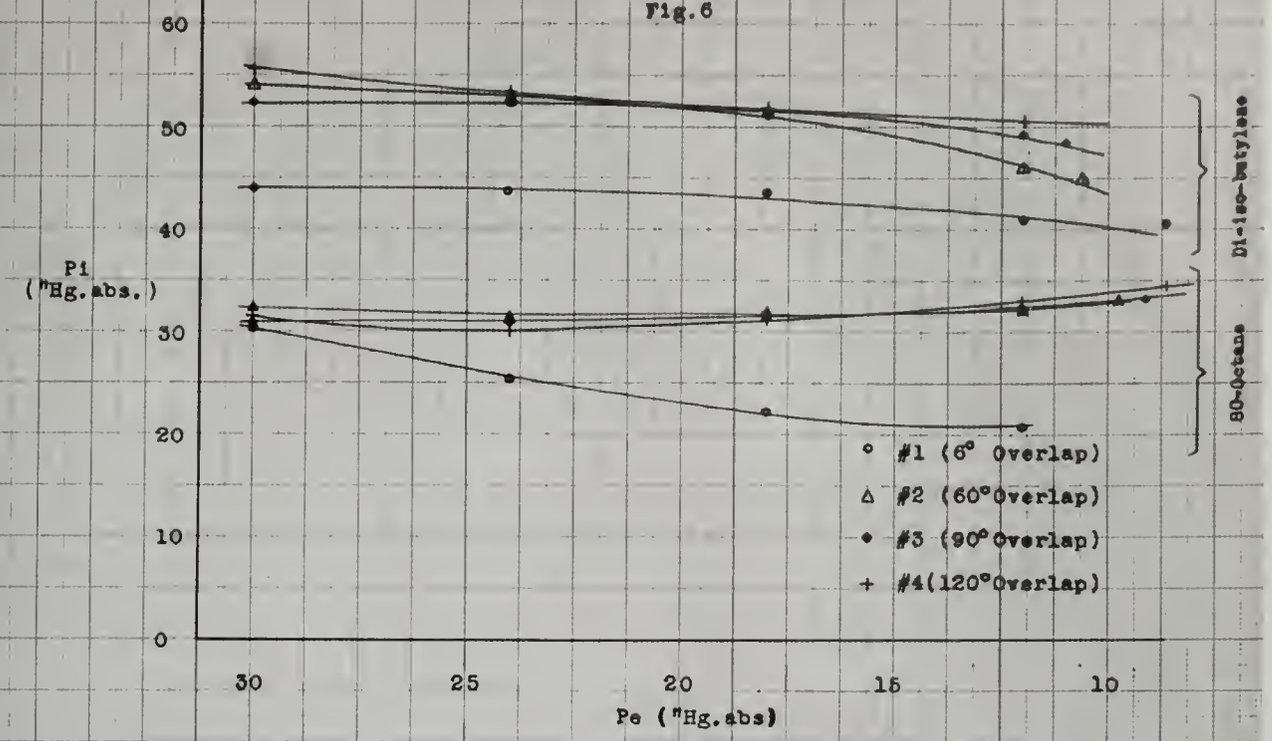
DI-ISO-BUTYLENE

Fig.5



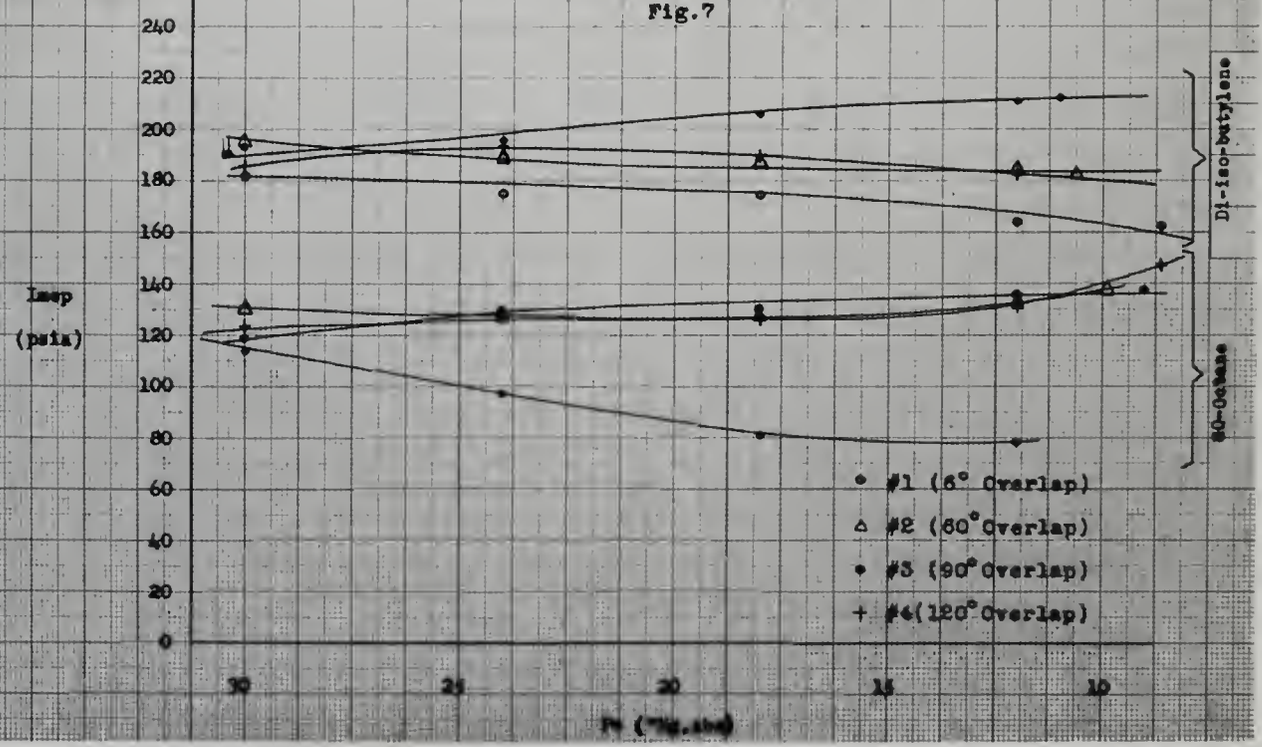
DETONATION LIMITED INLET PRESSURE

Fig.6



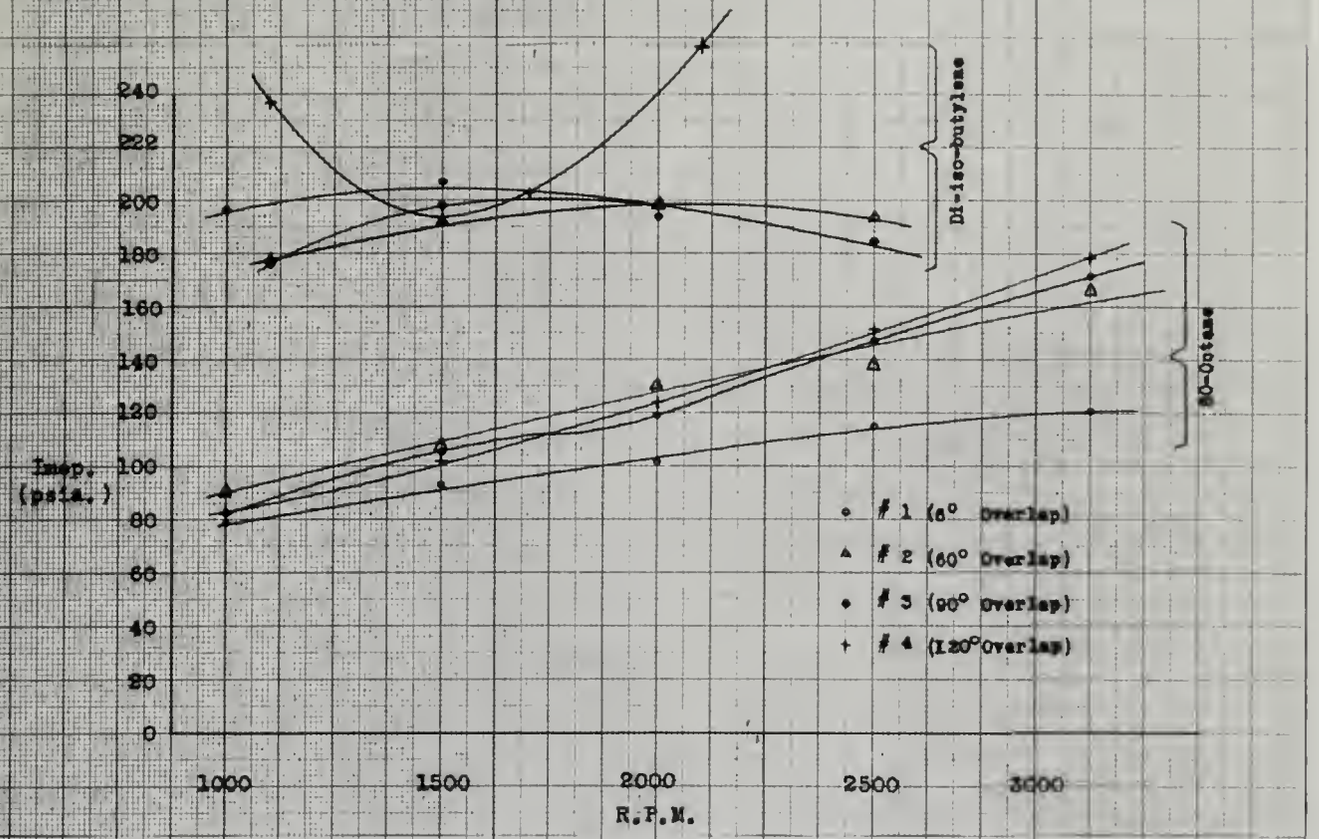
DETONATION LIMITED Imep

Fig.7



DETONATION LIMITED Imep.

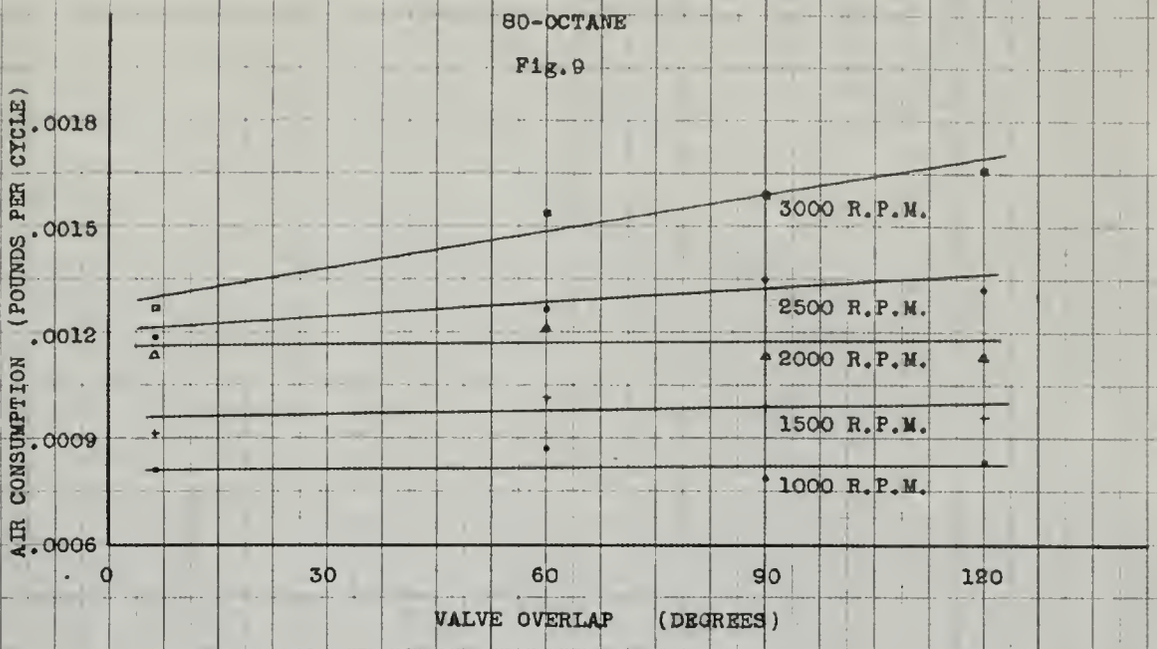
Fig. 8



AIR CONSUMPTION PER CYCLE

80-OCTANE

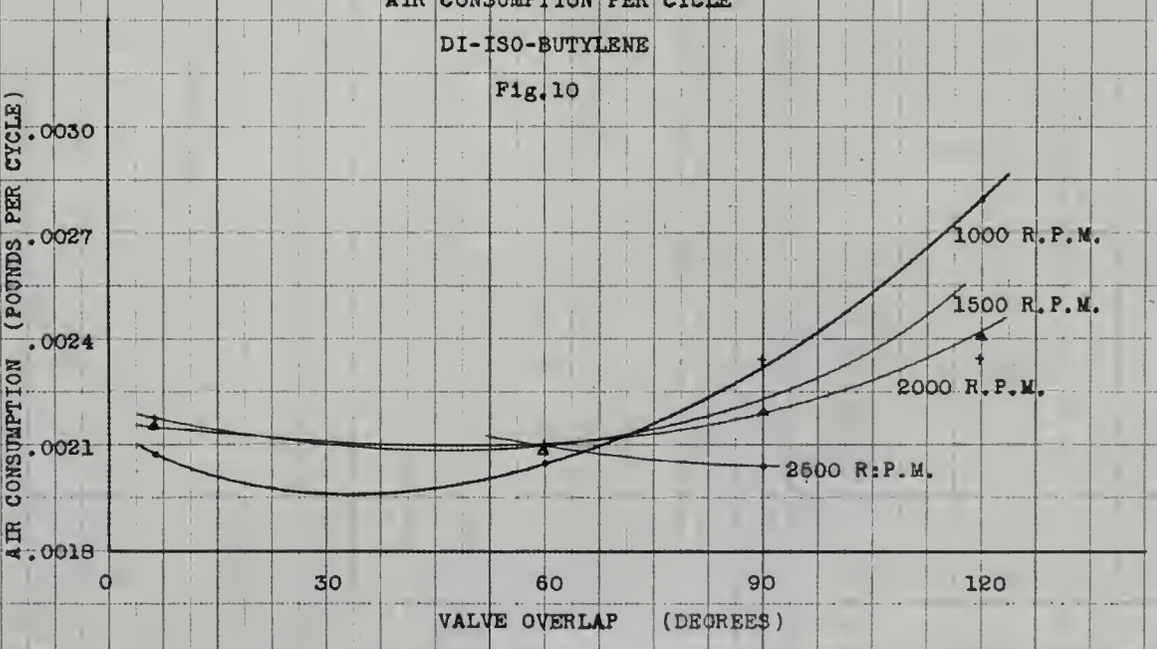
Fig.9



AIR CONSUMPTION PER CYCLE

DI-ISO-BUTYLENE

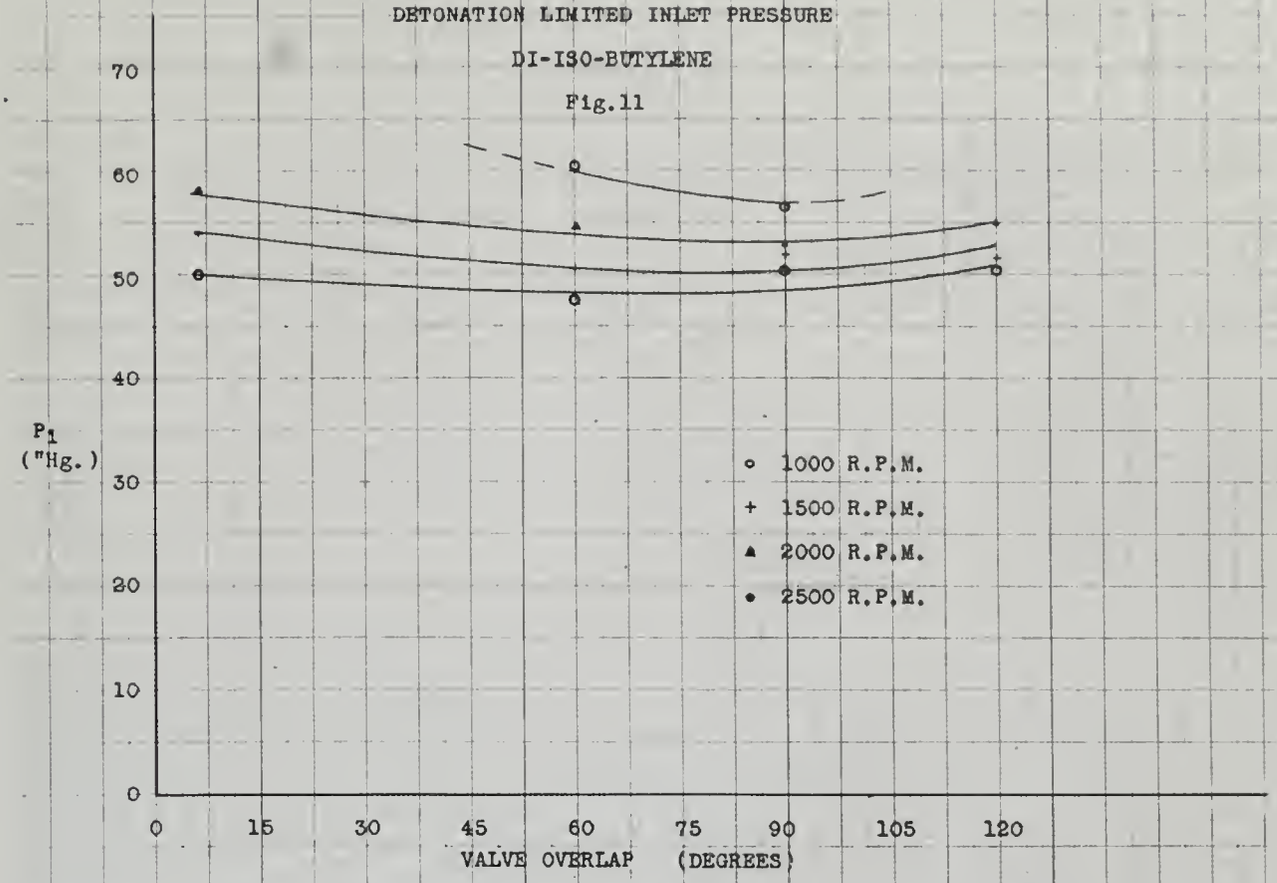
Fig.10



DETONATION LIMITED INLET PRESSURE

DI-ISO-BUTYLENE

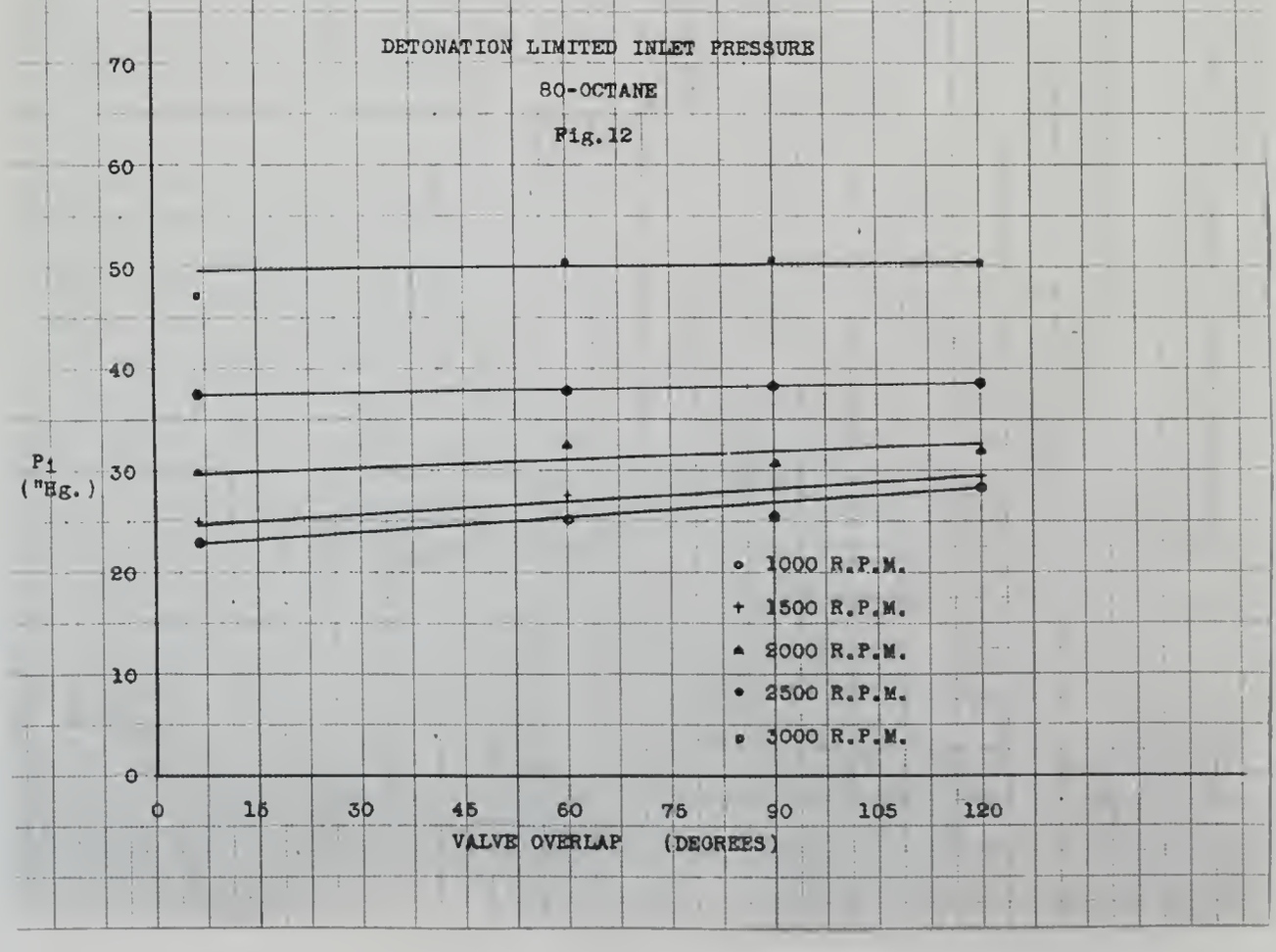
Fig.11



DETONATION LIMITED INLET PRESSURE

80-OCTANE

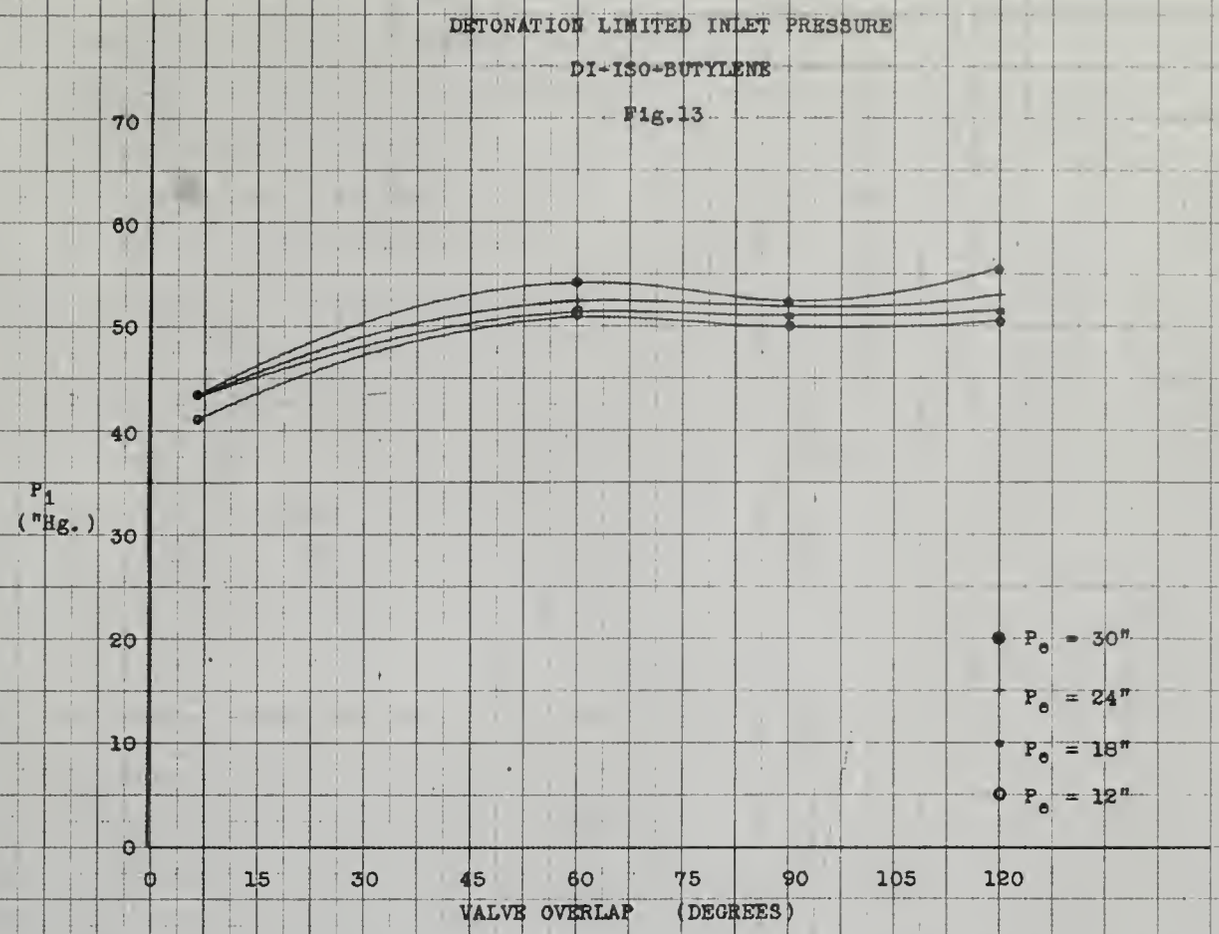
Fig.12



DETONATION LIMITED INLET PRESSURE

DI-ISO-BUTYLENE

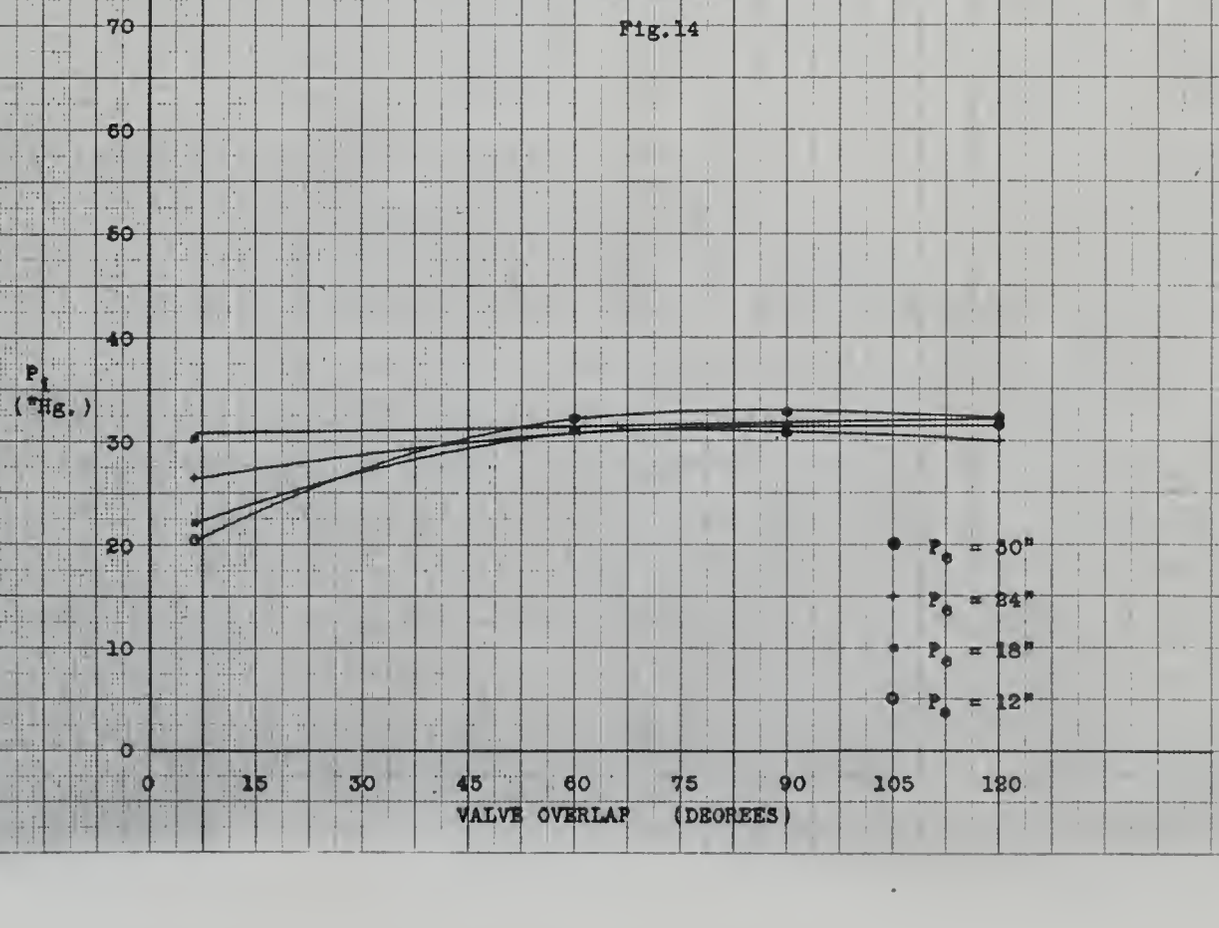
Fig.13



DETONATION LIMITED INLET PRESSURE

80-OCTANE

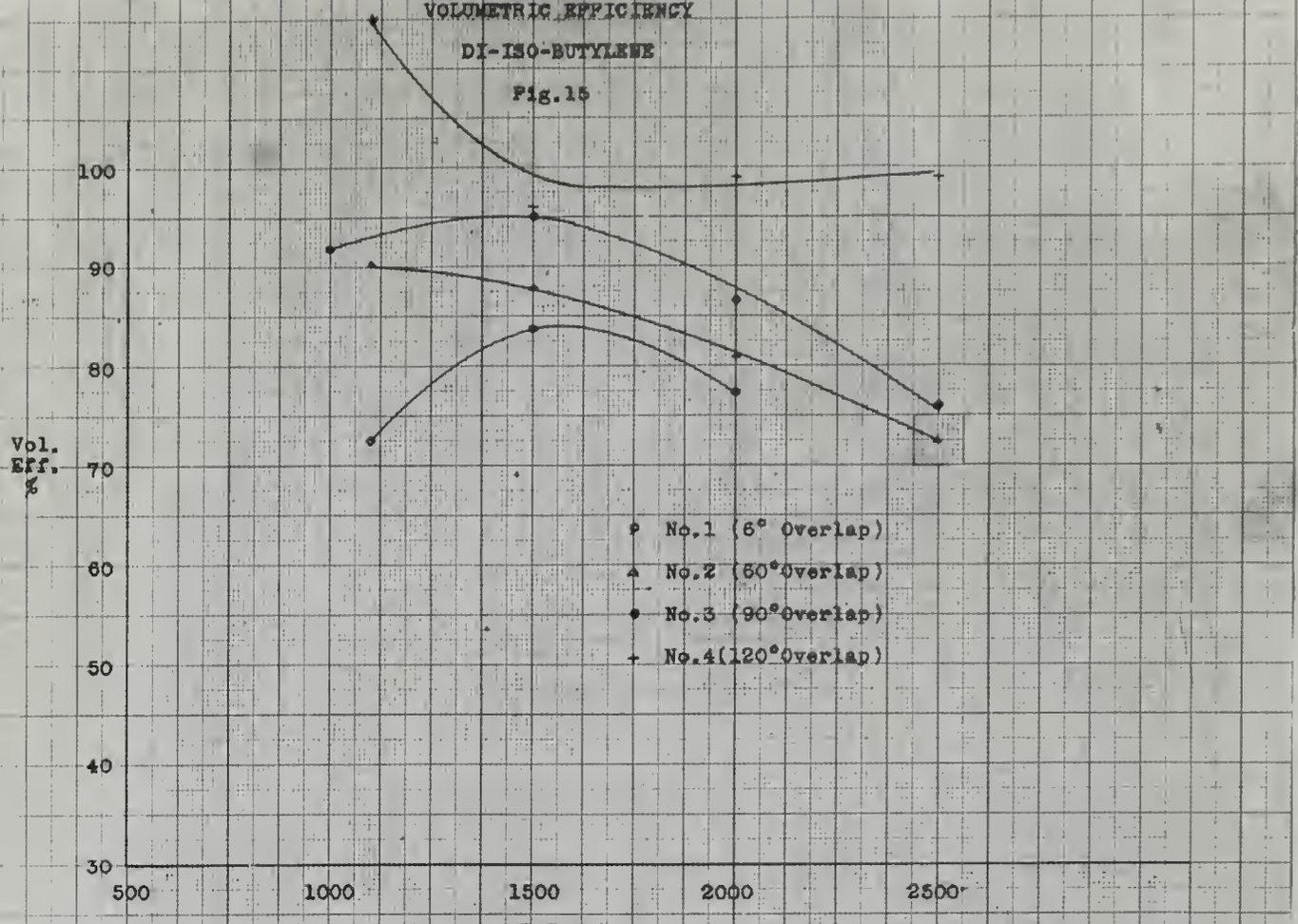
Fig.14



VOLUMETRIC EFFICIENCY

DI-ISO-BUTYLENE

Fig.15



VOLUMETRIC EFFICIENCY

80-OCTANE

Fig.16

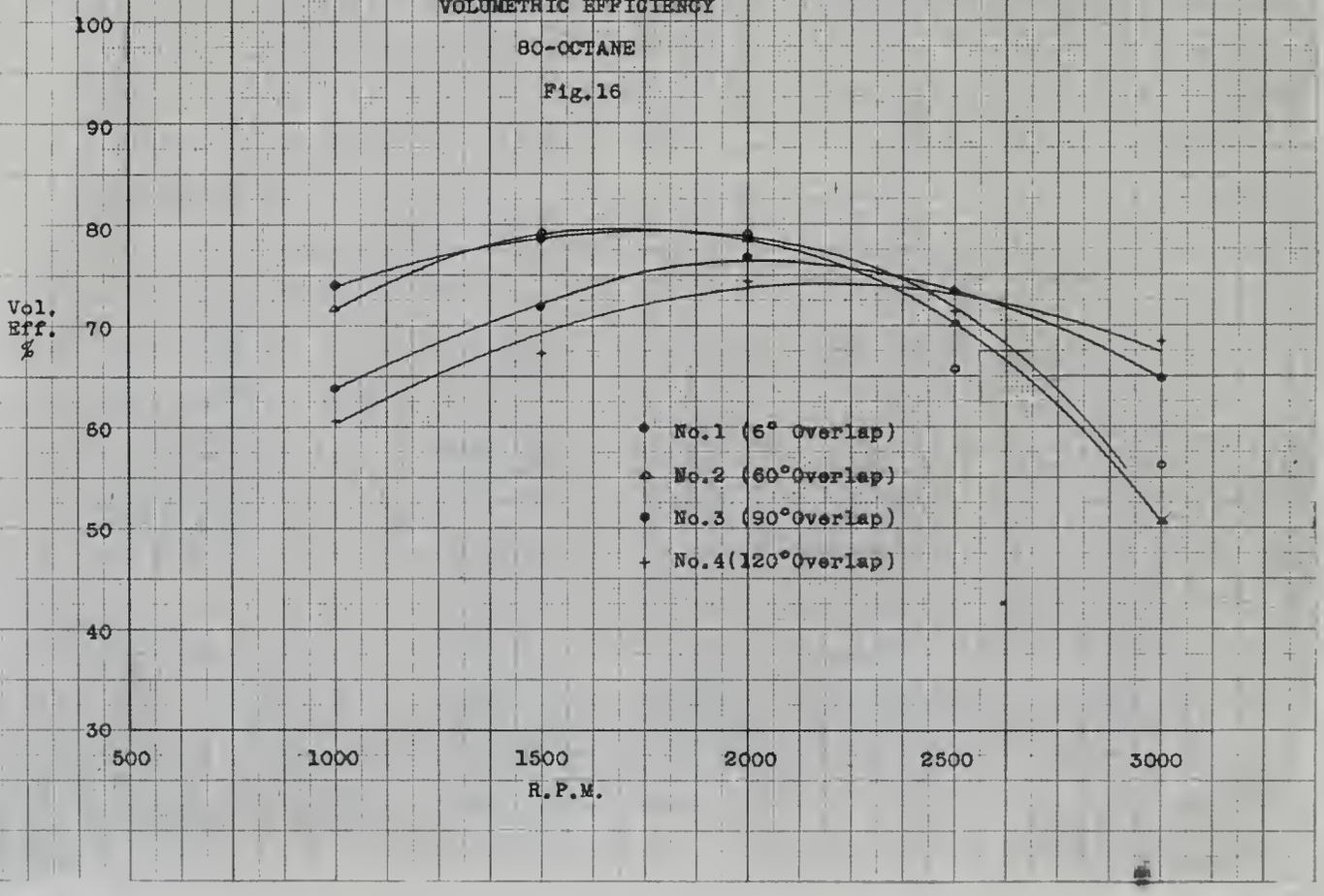


TABLE I

Shaft Number	1	Oil Temperature	140° F
Overlap	6°	Jacket Temperature	212° F
Compression Ratio	7	Inlet Temperature	140° F
Spark Advance	18° BTC	Fuel-Air Ratio	.08

RUN	RPM	B.L.	F.L.	P_i	P_e	ORF.	AIR CONS.	FUEL CONS.	R	ROTO.	T_b	ΔP	P_b	B.L. + F.L.	IHP	IMEP	ISFC
								80 Octane									
1	1000	14.25	-5.15	22.98	30.00	.515	.00675	.000540	1	8.0	77	1.5	52.33	19.4	3.88	82.3	.502
2	1500	15.4	-6.45	24.73	30.00	.515	.01138	.000910	1	10.6	77	4.1	50.98	21.85	6.56	92.7	.500
3	2000	16.6	-7.25	29.98	30.00	.515	.01890	.001515	1	14.3	77	12.0	48.13	23.85	9.54	101.2	.572
4	2500	19.05	-8.05	37.23	30.00	.725	.02435	.001948	2	9.1	77	4.95	48.23	27.1	13.55	115.0	.517
5	3000	19.3	-9.1	47.23	30.00	.725	.03170	.002540	2	10.78	77	7.45	54.43	28.4	17.04	120.5	.537
6	2000	19.95	-6.9	30.43	30.00	.515	.01767	.001412	1	13.7	72	10.35	48.28	26.85	10.74	114.0	.473
7	2000	16.85	-6.0	26.43	24.00	.515	.01515	.001212	1	12.55	72	7.2	49.23	22.85	9.14	97.0	.478
8	2000	13.2	-5.75	22.03	18.00	.515	.01338	.001070	1	11.6	72	5.75	49.68	18.95	7.58	80.4	.508
9	2000	12.9	-5.3	20.68	12.00	.515	.01198	.000958	1	10.9	72	4.6	49.93	18.2	7.28	77.2	.474
								Di-iso-butylene									
10	1100	7.7	-3.8	50.98	30.00	.515	.01900	.001520	1	14.8		9.15	63.08	41.4	9.1	175.8	.602
11	1500	11.8	-4.9	54.28	30.00	.515	.02710	.002170	1	18.6		19.5	59.93	46.6	13.98	198.0	.560
12	2000	10.6	-6.1	58.18	30.00	.725	.03580	.002860	2	12.0		8.0	64.48	46.6	18.62	198.0	.552
13	2500	-	-	70.00	30.00	.725	Incipient	Detonation		Not		Reached					
14	2000	36.2	-6.6	43.95	30.00	.515	.02800	.002240	1	19.0	77	23.4	54.01	42.8	17.13	182.0	.471
15	2000	35.2	-6.0	43.9	24.00	.515	.02665	.002130	1	18.4	77	19.7	57.70	41.2	16.50	175.0	.465
16	2000	35.5	-5.5	43.6	18.00	.515	.02610	.002085	1	18.1	77	19.0	57.50	41.0	16.40	174.0	.458
17	2000	33.7	-4.9	41.0	12.00	.515	.02480	.001980	1	17.55	77	17.0	58.30	38.6	15.44	163.8	.462
18	2000	33.3	-5.0	40.8	8.60	.515	.02450	.001960	1	17.5	77	17.2	57.80	38.3	15.31	162.4	.462

TABLE II

Shaft Number	2	Oil Temperature	140° F
Overlap	60°	Jacket Temperature	212° F
Compression Ratio	7	Inlet Temperature	140° F
Spark Advance	18° BTC	Fuel-Air Ratio	.08

RUN	RPM	B.L.	F.L.	P_i	P_e	ORF.	AIR CONS.	FUEL CONS.	R	ROTO.	T_b	ΔP	P_b	B.L. + F.L.	IHP	IMEP	ISFC
								80 Octane									
19	1000	17.8	-3.5	25.55	30.00	.515	.00726	.000580	1	8.4	80	2.15	39.90	21.30	4.26	90.4	.491
20	1500	20.8	-4.5	26.70	30.00	.515	.01260	.001070	1	11.3	80	6.70	38.35	25.3	7.58	107.4	.508
21	2000	25.4	-5.4	32.30	30.00	.515	.02020	.001615	1	14.9	80	13.00	50.80	30.8	12.31	130.8	.472
22	2500	26.7	-6.1	37.9	30.00	.725	.02640	.002110	2	9.5	80	5.95	47.45	32.8	16.40	139.0	.463
23	3000	31.9	-7.4	50.8	30.00	.725	.03840	.003070	2	12.3	80	10.50	56.75	39.3	23.60	166.8	.468
24	2000	25.4	-5.5	32.2	30.00	.515	.02020	.001616	1	14.9	80	13.10	50.80	30.9	12.36	131.0	.471
25	2000	25.6	-4.3	31.0	24.00	.515	.02000	.001600	1	14.8	80	10.00	64.95	29.9	11.98	126.9	.481
26	2000	26.7	-3.5	31.0	18.00	.515	.02030	.001620	1	15.0	80	10.25	65.10	30.2	12.08	128.0	.483
27	2000	28.8	-2.6	32.5	12.00	.515	.02090	.001670	1	15.2	80	11.70	64.00	31.4	12.55	133.1	.479
28	2000	29.9	-2.5	33.1	9.50	.515	.02210	.001770	1	15.7	80	12.35	63.95	32.4	12.96	137.5	.492
								Du-iso-butylene									
29	1100	39.6	-2.4	47.33	30.00	.515	.01870	.001500	1	14.7	76	7.75	72.78	42.0	8.40	178.2	.642
30	1500	42.1	-3.0	50.23	30.00	.515	.02630	.002100	1	18.1	76	17.60	63.10	45.1	13.58	191.3	.557
31	2000	42.3	-4.7	54.23	30.00	.725	.03500	.002800	2	11.8	76	7.40	66.43	47.0	18.80	199.4	.536
32	2500	40.4	-5.5	60.63	30.00	.725	.04370	.003500	2	13.8	76	11.55	66.68	45.9	22.95	194.8	.550
33	3000	-	-	70.00	30.00	.725	Incipient	Detonation		Not	Reached						
34	2000	42.0	-4.2	54.23	30.00	.725	.03500	.002800	2	11.8	76	7.40	66.43	46.2	18.50	196.0	.545
35	2000	41.3	-3.2	52.73	24.00	.515	.03400	.002720	2	11.6	76	28.80	64.63	44.5	17.80	189.0	.550
36	2000	41.0	-3.1	51.63	18.00	.515	.03480	.002780	2	11.7	76	31.80	59.73	44.1	17.65	187.0	.566
37	2000	41.5	-3.3	50.93	12.00	.515	.03370	.002700	2	11.6	76	30.70	59.80	43.8	17.52	185.9	.554
38	2000	39.5	-3.5	49.83	10.60	.515	.03340	.002670	2	11.5	76	29.70	60.43	43.0	17.20	182.4	.559

TABLE III

Shaft Number	3	Oil Temperature	140° F
Overlap	90°	Jacket Temperature	212° F
Compression Ratio	7	Inlet Temperature	140° F
Spark Advance	18° BTC	Fuel-Air Ratio	.08

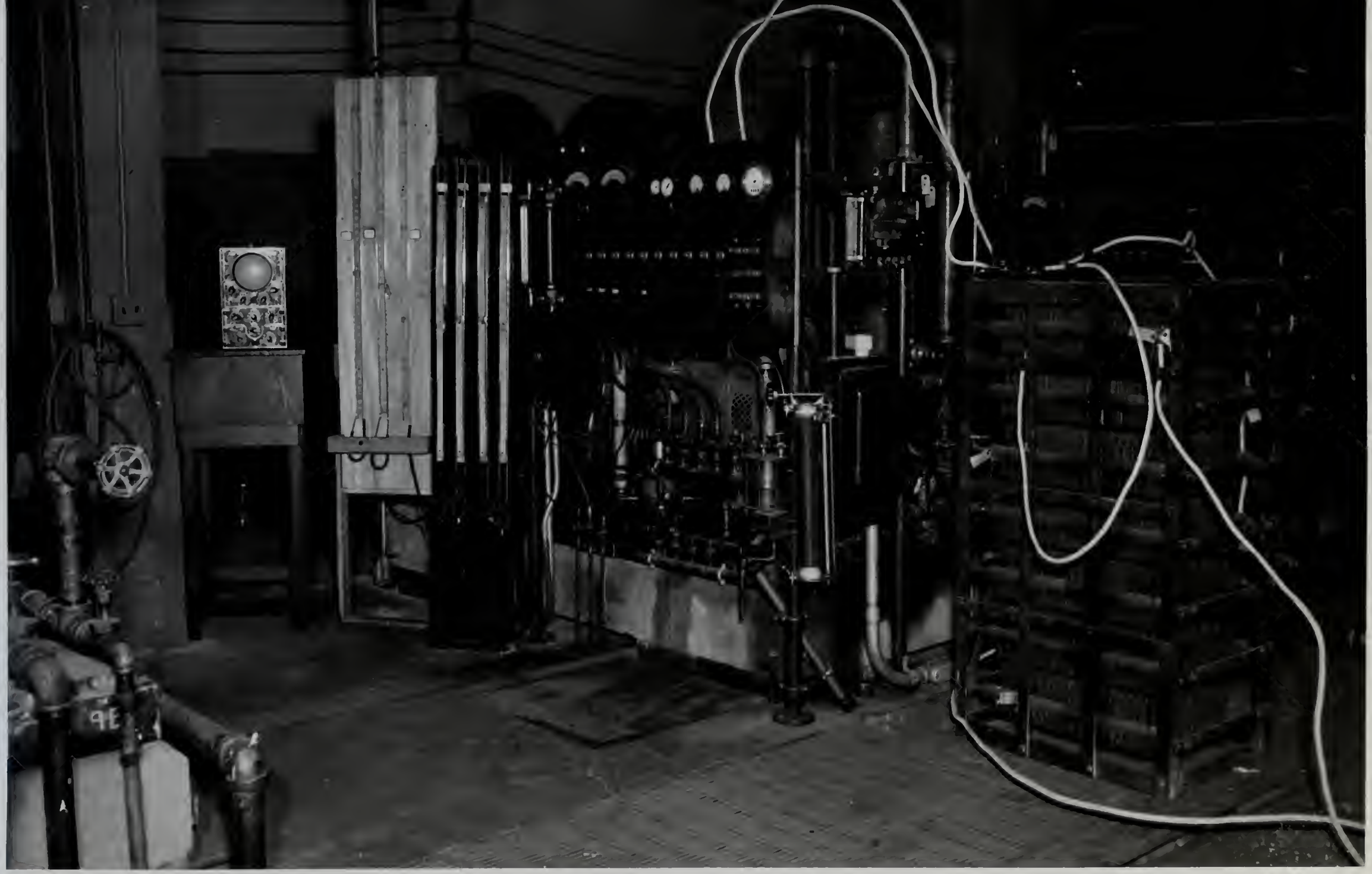
RUN	RPM	B.L.	F.L.	P_i	P_e	ORF.	AIR CONS.	FUEL CONS.	R	ROTO.	T_b	ΔP	P_b	B.L. +F.L.	IHP	IMEP	ISFC
								80 Octane									
39	1000	14.3	-4.2	25.77	30.00	.515	.00653	.000522	1	8.0	75	1.0	68.02	18.5	3.75	78.5	.501
40	1500	19.8	-4.9	28.52	30.00	.515	.01228	.000980	1	11.2	75	3.65	66.37	24.7	7.41	105.0	.476
41	2000	21.6	-6.5	30.77	30.00	.725	.01880	.001510	2	7.8	75	2.2	64.52	28.1	11.25	119.4	.482
42	2500	27.3	-7.3	38.42	30.00	.725	.02810	.002250	2	9.9	75	5.2	60.97	34.6	17.3	147.0	.468
43	3000	31.3	-8.6	51.2	30.00	.725	.03960	.003170	2	12.6	75	11.2	55.62	40.4	24.3	171.8	.469
44	2000	21.5	-6.5	30.77	30.00	.725	.01880	.001510	2	7.8	75	2.2	64.52	28.0	11.2	119.0	.486
45	2000	24.5	-5.8	30.92	24.00	.515	.02080	.001660	1	15.1	75	11.0	62.90	30.3	12.12	128.8	.492
46	2000	25.7	-5.0	31.42	18.00	.515	.02190	.001750	1	15.7	75	12.8	60.17	30.7	12.3	130.4	.512
47	2000	27.6	-4.4	32.42	12.00	.725	.02290	.001830	1	16.0	75	3.25	64.47	32.0	12.8	135.9	.514
48	2000	28.2	-4.1	33.02	9.20	.725	.02340	.001870	1	16.4	75	3.40	63.42	32.3	12.93	137.2	.521
								Di-iso-butylene									
49	1000	42.4	-3.7	50.3	30.00	.515	.02020	.001620	1	15.5	75	10.1	64.80	46.1	9.22	196.0	.632
50	1500	44.1	-4.8	51.5	30.00	.725	.02920	.002340	2	10.6	75	5.6	61.10	48.9	14.7	207.0	.573
51	2000	39.6	-6.0	52.82	30.00	.725	.03650	.002920	2	12.3	75	8.2	65.05	45.6	18.25	193.5	.577
52	2500	36.4	-7.1	56.3	30.00	.725	.04260	.003400	2	13.5	75	11.8	61.2	43.5	21.8	184.6	.561
53	3000	-	-	70.00	30.00	.725	Incipient	Detonation		Not		Reached					
54	2000	39.6	-6.0	52.3	30.00	.515	.03650	.002920	2	12.3	75	8.2	65.05	45.6	18.25	193.5	.577
55	2000	41.2	-4.9	52.0	24.00	.515	.03660	.002930	2	12.2	75	8.35	64.2	46.1	18.42	196.0	.572
56	2000	44.0	-4.5	51.0	18.00	.515	.03910	.003130	2	12.7	75	9.65	63.2	48.5	19.4	206.0	.612
57	2000	45.6	-4.1	49.0	12.00	.515	.04020	.003220	2	12.9	75	10.4	62.3	49.7	19.9	211.0	.597
58	2000	46.1	-4.0	48.5	11.00	.515	.04080	.003270	2	13.1	75	10.8	61.95	50.1	20.05	212.5	.587

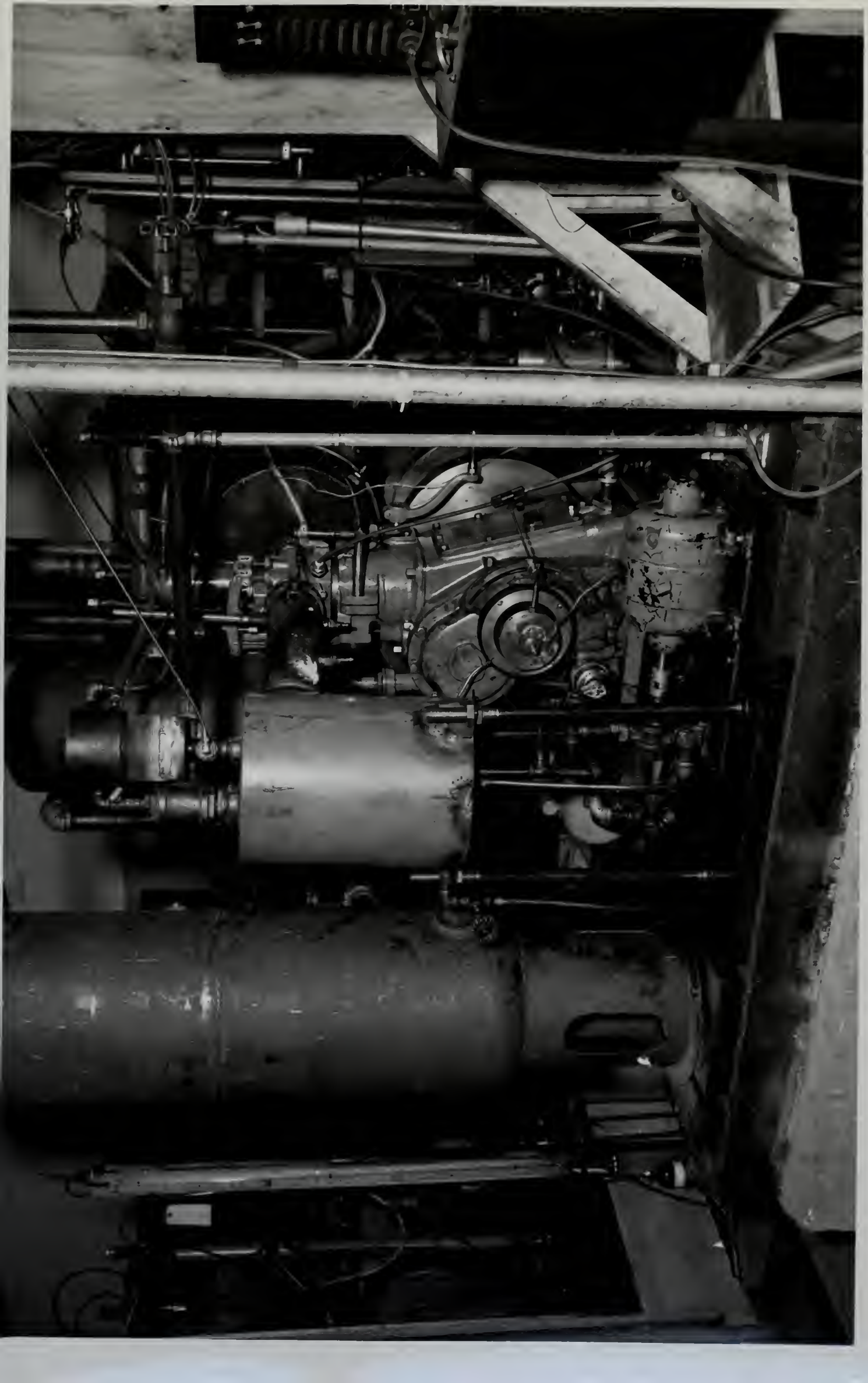
TABLE IV

Shaft Number 4
Overlap 120°
Compression Ratio 7
Spark Advance 18° BTC

Oil Temperature 140° F
Jacket Temperature 212° F
Inlet Temperature 140° F
Fuel-Air Ratio .08

RUN	RPM	B.L.	F.L.	P_i	P_e	ORF.	AIR CONS.	FUEL CONS.	R	ROTO.	T_b	ΔP	P_b	B.L. +F.L.	IHP	IMEP	ISFC
								80 Octane									
60	1000	15.5	-4.1	28.55	30.00	.515	.00689	.000543	1	8.1	70	1.15	63.75	19.60	3.92	83.2	.407
61	1500	18.6	-5.4	29.35	30.00	.515	.01180	.000935	1	10.7	70	3.50	62.15	24.00	7.2	101.9	.468
62	2000	22.8	-6.4	31.55	30.00	.515	.01870	.001498	1	14.2	70	9.35	60.00	29.2	11.68	124.0	.462
63	2500	27.9	-7.6	38.55	30.00	.515	.02750	.002200	1	18.2	70	21.25	56.80	35.5	17.75	150.8	.446
64	3000	33.6	-8.4	50.85	30.00	.725	.04160	.003330	2	13.0	70	12.60	54.65	42.0	25.24	178.2	.475
65	2000	22.8	-6.2	31.50	30.00	.515	.01870	.001500	1	14.2	70	9.35	60.00	29.0	11.6	123.0	.466
66	2000	24.4	-5.6	30.00	24.00	.515	.02030	.001625	1	15.0	70	9.90	65.65	30.0	12.0	127.2	.488
67	2000	24.5	-5.1	31.25	18.00	.515	.02280	.001820	1	16.1	70	12.75	64.90	29.6	11.84	125.5	.553
68	2000	26.3	-4.4	33.05	12.00	.515	.02480	.001970	1	16.9	70	15.05	64.20	30.7	12.28	130.2	.579
69	2000	30.8	-3.8	34.35	8.60	.515	.02600	.002080	1	17.5	70	17.0	63.35	34.6	13.84	146.9	.541
								Di-iso-butylene									
70	1100	51.9	-4.0	50.86	30.00	.515	.02565	.002050	2	9.7	75	16.55	63.71	55.9	12.30	237.0	.600
71	1500	40.2	-5.4	51.00	30.00	.515	.02930	.002350	2	10.6	78	22.20	62.51	45.6	13.67	193.5	.619
72	1700	42.8	-5.1	51.21	30.00	.515	.03440	.002750	2	11.7	78	31.3	60.81	47.9	16.30	203.0	.607
73	2100	55.2	-6.0	56.0	30.00	.725	.04200	.003380	2	13.4	75	11.05	63.65	61.2	25.67	259.5	.473
74	2500	-	-	70.00	30.00	.725	Incipient	Detonation		Not		Reached					
75	2000	37.8	-6.1	55.6	30.00	.725	.04180	.003350	2	13.3	78	11.03	63.60	43.9	17.56	186.3	.687
76	2000	40.3	-5.1	53.1	24.00	.725	.04060	.003250	2	13.0	78	10.55	63.20	45.4	18.18	192.9	.644
77	2000	40.1	-4.5	51.5	18.00	.725	.03970	.003180	2	12.8	78	9.80	64.60	44.6	17.80	189.1	.643
78	2000	38.9	-4.4	50.3	12.00	.725	.03810	.003050	2	12.5	78	9.00	65.00	43.3	17.33	183.8	.632





DATE DUE

Thesis
M184 McElroy 8094
The effect of valve overlap on detonation limited engine performance.

Thesis 8094
M184 McElroy
The effect of valve overlap on detonation limited engine performance.

thesM184
The effect of valve overlap on detonatio
3 2768 001 88428 1
DUDLEY KNOX LIBRARY